

Chapter 2.3

Platforms with Multi-directional Total Thrust

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Abstract The chapter provides an overview of the basic modeling and the intrinsic properties of aerial platforms with multi-directional total thrust ability. When also fully-actuated, such platforms can modify the total wrench in body frame in any direction, thus allowing the control of position and orientation independently. Therefore, they are best suited for dexterous tasks, physical interaction, and for carrying aerial manipulators, because they do not suffer from the underactuation of standard collinear multirotors. The chapter includes a rigorous classification, a discussion on the possible input coupling, and on the capabilities and pitfalls of inverse-dynamics control approach for such platforms.

1 Introduction and Benefits of Multi-directional Total Thrust

Aerial vehicles have been thoroughly studied and applied in several fields and for several tasks, from simple remote sensing to the more challenging physical interaction with the environment and humans. The latter have been firstly targeted using aerial vehicles actuated by *multiple collinear rotors* and endowed with cables [289, 84, 291] rigid tools [207, 315] or a more complex robotic arm with a few degrees of freedom (DoFs) [88, 200, 292]. Collinear-rotor vehicles are energy efficient but underactuated because of the unidirectionality of the total thrust in the body frame. As a consequence the vehicle orientation is coupled with its translational motion, and the vehicle cannot *instantaneously* react to forces with any direction.

Cases in which a tool is rigidly fixed to the airframe have been presented in [101, 15, 317, 207, 100]. The impossibility of controlling the 6D (position plus orientation) dynamics of the end-effector limits the potential use cases and also creates stability issues. In fact, it has been shown that in the presence of interactions with points of the airframe other than the vehicle center of mass (CoM) the internal dynamics of underactuated multirotors is not guaranteed to be stable, and it is, in general, neither easy to stabilize nor practical for real applications [207].



Fig. 38 The TiltHex (left) and the OTHex (right): two fully actuated hexarotors by LAAS-CNRS. They are capable of 6D motion (fully-decoupled pose) and 6D physical interaction thanks to the non-collinear orientations of the thrust forces produced by its propellers and the use of suitable physical interfaces, such as the passive gripper of the OTHex, optimized for bar lifting.

To overcome such limitation, the main approach has been to attach an n -DoF robotic arm to the aerial platform [200, 199, 18, 138, 292], a solution which aims at overcoming the underactuation of the end-effector dynamics exploiting the increased number of actuators provided by the arm. In this way, a fully actuated 6D force control at the end-effector side becomes possible [313]. However, this solution comes with a few drawbacks as well, the main being that: *i*) a robotic arm strongly decreases the payload and flight time due to its own weight; *ii*) the system is much more complex from a mechanical point of view than a single airframe with a rigid tool and, thus, it is more expensive to build and also requires more maintenance and repairing costs across its operational life; *iii*) lateral forces in body frame, which cannot be provided by the aerial platform itself, have to be generated through the dynamical/inertial coupling between the arm and the aerial robot: the proper mastering of the dynamical coupling is something that has to be necessarily exploited in order to get the sought benefits in terms of 6D force control. This, in turn, requires the knowledge of the precise dynamical model and a very accurate measurement of the system inputs and states (position, orientation, linear and angular velocities). As a matter of fact, these requirements are extremely hard to achieve in real world conditions (especially the former). For this reason, kinematic-only approaches have been preferred for real world validations, see e.g., [200, 199], at the expense of losing the main benefits for which the manipulator was introduced.

Summarizing, standard flying platforms are underactuated and, thus, incapable of 6D end-effector force control. On the other side, to bring a full manipulator up to the air to perform the sought 6D end-effector force control is often excessively complex and may introduce more problems than benefits, depending on the task. To solve all these problems at once and finally achieve the sought full 6D force control of the aerial interaction, is instead enough to let the aerial vehicle possess the minimal requirements to perform such interaction with a rigidly attached end-effector. Such requirements can be satisfied with the use of a multirotor with *generically-oriented fixed propellers* (GOFP) instead of the more common *collinearly-oriented fixed propeller* (COFP) architectures [254]. In GOFP platforms, which appeared in the robotics literature only recently (see, e.g., [237, 306, 244, 41, 225]), the full-actuation is achieved by a more general propeller position and orientation. The dif-

ference between the underactuated platform and the last approach is that, in the former approach, all the propellers have the same orientation while, in the second approach, every propeller orientation is different. The latter approach is, thus, able to control independently the translational and angular acceleration in contact-free flight, or any of the six components of the exerted wrench when in contact, thus allowing full and dexterous 6D force control, which makes them much more suited for physical interaction tasks than standard COFP platforms. An example of a fully actuated GOFP platform is the OTHex by LAAS-CNRS, depicted in Figure 38. This platform has six propellers with coplanar centers and non-collinear directions of rotation, a configuration which makes it possible the 6D physical interaction and manipulation [280].

Another solution to obtain full-actuation consists of actively tilting the whole propeller groups [254, 255, 170], a solution which is called thrust vectoring or tilting propeller. This solution however is subject to the same drawbacks of the solutions employing a manipulator arm, since they require extra actuation, mechanical complexity, and weight. Furthermore, they cannot in general guarantee instantaneous disturbance rejection or fast force exertion since the propellers might have to be re-oriented, which again takes some non-negligible time.

In this Chapter we introduce a basic but effective model of GOFP platforms and provide a major classification and properties of such vehicles. In particular, the model of the total thrust and moment generation, and classification with respect to actuation singularity are provided in Sec. 2. In Sec. 3 we describe the capability of producing a multi- or omnidirectional total thrust when employing propellers with uni-directional lift force. The possible coupling between total thrust and total moment is described in Sec. 4. Sec. 5 introduces the inverse dynamics control strategy for fully actuated platforms and highlights its limits, which will be overcome in Chapter 3.6. Finally, conclusions are drawn in Sec. 6.

2 Platforms with Generically-Oriented Fixed Propellers

Let us define an inertial world frame $\mathcal{F}_W = \{O_W, \mathbf{x}_W, \mathbf{y}_W, \mathbf{z}_W\}$ where O_W is its origin, placed arbitrarily, and $(\mathbf{x}_W, \mathbf{y}_W, \mathbf{z}_W)$ are the orthogonal unit vectors. We consider \mathbf{z}_W parallel and opposite to the gravity vector. Then we define the body frame $\mathcal{F}_B = \{O_B, \mathbf{x}_B, \mathbf{y}_B, \mathbf{z}_B\}$ rigidly attached to the vehicle and centered in O_B , the vehicle center of mass (CoM). The position of O_B and orientation of \mathcal{F}_B w.r.t. \mathcal{F}_W are described by the vector $\mathbf{p}_b \in \mathbb{R}^3$ and the rotation matrix $\mathbf{R}_b \in SO(3)$, respectively. Then we define by the vector $\mathbf{v}_b \in \mathbb{R}^3$ the translational velocity of O_B expressed in \mathcal{F}_W , and by $\boldsymbol{\omega}_b^b \in \mathbb{R}^3$ the angular velocity of \mathcal{F}_B w.r.t. \mathcal{F}_W and expressed in \mathcal{F}_B . The generic vehicle, with only 3 of the n propellers, is depicted in Fig. 112.

The vehicle is modeled as a rigid body with mass $m \in \mathbb{R}_{>0}$ and moment of inertia about O_B , defined w.r.t. \mathcal{F}_B , described by the positive definite matrix $\mathbf{I}_b \in \mathbb{R}^{3 \times 3}$. The dynamics of the system is computed applying the Newton-Euler equations, thus obtaining $\dot{\mathbf{p}}_b = \mathbf{v}_b$, $\dot{\mathbf{R}}_b = \mathbf{R}_b \boldsymbol{\Omega}_b$, and

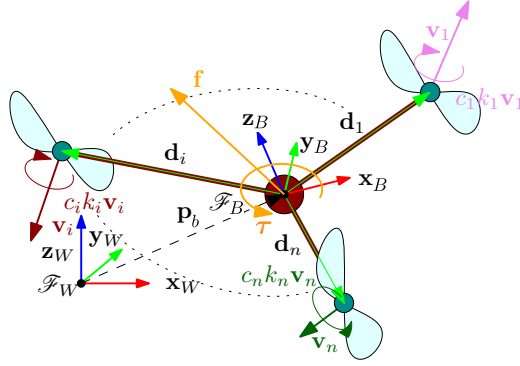


Fig. 39 Schematic representation of a multirotor with generically oriented fixed propellers and its main quantities. Only three of the n propellers are shown.

$$\underbrace{\begin{bmatrix} m\mathbf{I}_3 & \mathbf{0} \\ \mathbf{0} & \mathbf{I}_b \end{bmatrix}}_{\mathbf{M}_R} \underbrace{\begin{bmatrix} \dot{\mathbf{v}}_b \\ \dot{\boldsymbol{\omega}}_b^b \end{bmatrix}}_{\mathbf{a}} = \underbrace{\begin{bmatrix} -g\mathbf{e}_3 \\ -\mathbf{I}_b \boldsymbol{\omega}_b^b \times \boldsymbol{\omega}_b^b \end{bmatrix}}_{\mathbf{b}_R} + \underbrace{\begin{bmatrix} \mathbf{R}_b & \mathbf{0} \\ \mathbf{0} & \mathbf{I}_3 \end{bmatrix}}_{\mathbf{G}_R} \begin{bmatrix} \mathbf{f}_u^b \\ \boldsymbol{\tau}_u^b \end{bmatrix}, \quad (40)$$

where $\mathbf{e}_3 = [0 \ 0 \ 1]^\top$, $\boldsymbol{\Omega}_b = \mathbf{S}(\boldsymbol{\omega}_b^b)$ is the skew symmetric matrix relative to $\boldsymbol{\omega}_b^b$, $\mathbf{f}_u^b \in \mathbb{R}^3$ and $\boldsymbol{\tau}_u^b \in \mathbb{R}^3$ are the controllable total input force and torque expressed in \mathcal{F}_B , respectively.

Considering a multirotor with n rotors, each of them produces a lift force and a moment due to the drag force [299]. All together they generate the total force (or thrust) and moment, \mathbf{f} and $\boldsymbol{\tau}$, respectively,¹ expressed as:

$$\mathbf{w} = [\mathbf{f}^\top \ \boldsymbol{\tau}^\top]^\top = [\mathbf{G}_1^\top \ \mathbf{G}_2^\top]^\top [u_1 \ \dots \ u_n]^\top = \mathbf{G}\mathbf{u}. \quad (41)$$

The matrixes $\mathbf{G} \in \mathbb{R}^{6 \times n}$, $\mathbf{G}_1 \in \mathbb{R}^{3 \times n}$, and $\mathbf{G}_2 \in \mathbb{R}^{3 \times n}$ are called the *full allocation matrix*, the *force allocation matrix* and the *moment allocation matrix*, respectively. The control $u_i \in \mathbb{R}$ is typically equal to $\omega_i |\omega_i|$, where $\omega_i \in \mathbb{R}$ is the i -th propeller rotational speed. \mathbf{G}_1 and \mathbf{G}_2 have the following structure

$$\mathbf{G}_1 = [\mathbf{v}_1 \ \dots \ \mathbf{v}_n], \quad (42)$$

$$\mathbf{G}_2 = [\mathbf{d}_1 \times \mathbf{v}_1 \ \dots \ \mathbf{d}_n \times \mathbf{v}_n] + [c_1 k_1 \mathbf{v}_1 \ \dots \ c_n k_n \mathbf{v}_n], \quad (43)$$

where i) $\mathbf{v}_i \in \mathbb{R}^3$ are the coordinates, in \mathcal{F}_B , of the lift force generated by the i -th propeller when $u_i = 1$. In this formulation the aerodynamic coefficient that maps propeller speed into thrust intensity, typically called lift factor c_T , is $c_{Ti} = \|\mathbf{v}_i\| = v_i$; ii) \mathbf{d}_i is the position of the center of the i -th propeller in body frame; iii) $c_i = -1$ ($c_i = 1$) if the i -th propeller angular velocity vector has the same direction of \mathbf{v}_i ($-\mathbf{v}_i$) when $u_i > 0$, i.e., the propeller spins CCW (CW) when watched from its top;

¹ In the following we omit the subscripts and superscripts when there is no risk of confusion.

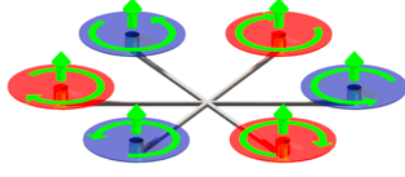


Fig. 40 A standard underactuated collinear hexarotor.

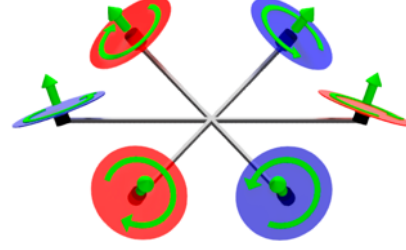


Fig. 41 The fully actuated non-collinear hexarotor called 'TiltHex', by LAAS-CNRS.

iv) $k_i \in \mathbb{R}$ is the constant ratio between the i -th propeller lift force and the drag moment, typically denoted with c_Q/c_T in the literature. The following is a well known and easy-to-prove fact.

Fact 1 (translation invariance) \mathbf{G} does not change if \mathbf{d}_i is replaced with $\mathbf{d}_i + \lambda_i \mathbf{v}_i$ for any $i = 1, \dots, n$ and $\lambda_1, \dots, \lambda_n \in \mathbb{R}$.

We introduce now the basic concept of multirotor design. Let us first define $\mathbf{c} = [c_1 \dots c_n]^\top$ and $\mathbf{k} = [k_1 \dots k_n]^\top$.

Definition 1 A multirotor design is a tuple $\mathcal{D} = (n, \mathbf{c}, \mathbf{k}, \mathbf{d}_1, \dots, \mathbf{d}_n, \mathbf{v}_1, \dots, \mathbf{v}_n)$, which describes the number of propellers n , their aerodynamic characteristics, locations and orientations w.r.t. \mathcal{F}_B . We call the tuples $(\mathbf{v}_1, \dots, \mathbf{v}_n)$ and $(n, \mathbf{c}, \mathbf{k}, \mathbf{d}_1, \dots, \mathbf{d}_n)$ the vectoring part and the etero-vectoring part of \mathcal{D} , respectively.

A first important classification of multirotor design is the following

Definition 2 A multirotor design is:

- Underactuated if $\text{rank}(\mathbf{G}) < 6$
- Fully actuated if $\text{rank}(\mathbf{G}) = 6$

furthermore, the multirotor translational (resp. rotational) dynamics are

- Underactuated if $\text{rank}(\mathbf{G}_1) < 3$ (resp. $\text{rank}(\mathbf{G}_2) < 3$)
- Fully actuated if $\text{rank}(\mathbf{G}_1) = 3$ (resp. $\text{rank}(\mathbf{G}_2) = 3$).

Full actuation requires $n \geq 6$, which is however, only a necessary condition. In fact, classical multirotor designs, like the one depicted in Fig. 40 have the $\mathbf{v}_1, \dots, \mathbf{v}_n$ all aligned and therefore $\text{rank}(\mathbf{G}_1) = 1$, which means that their translational dynamics is underactuated. On the other hand they are such that $\text{rank}(\mathbf{G}_2) = 3$, i.e., the rotational dynamics is fully actuated, and $\text{rank}(\mathbf{G}) = 4$. Other designs, such as the one depicted in Fig. 41, named TiltHex, by LAAS-CNRS, and presented in [237], has the $\mathbf{v}_1, \dots, \mathbf{v}_n$ non collinear and such that $\text{rank}(\mathbf{G}) = 6$ which makes it a fully actuated platform.

Notice that choosing non-collinear \mathbf{v}_i 's is a necessary but non-sufficient condition to obtain full actuation, in fact one has to ensure that all the 6 row vectors of \mathbf{G} are linearly independent.

3 Multi-directional and Omnidirectional Total Thrust Platforms

We denote with $\mathbf{1}$ the column vector with all ones. Its size is understood from the context. Given two vectors \mathbf{x} and \mathbf{y} , the notations $\mathbf{x} \geq \mathbf{y}$, $\mathbf{x} > \mathbf{y}$ have to be intended component-wise. Accordingly, we define $\mathbb{R}_{\geq \mathbf{0}}^n = \{\mathbf{x} \in \mathbb{R}^n | \mathbf{x} \geq \mathbf{0}\}$ and $\mathbb{R}_{> \mathbf{0}}^n = \{\mathbf{x} \in \mathbb{R}^n | \mathbf{x} > \mathbf{0}\}$, called the non-negative orthant and the positive orthant of \mathbb{R}^n , respectively.

Propellers can be of mainly two types: mono-directional and bidirectional. Mono-directional thrusters are the cheapest and most common solution but can produce lift only in one direction, a constraint which can be encoded imposing $\mathbf{u} \in \mathbb{R}_{\geq \mathbf{0}}^n$ in (41). Bidirectional thrust rotors are able to invert the direction of the lift force by inverting either the motor rotation or the propeller angle of attack. However such rotors have several issues: i) scarceness of reversible Electronic Speed Controllers (ESC) for brushless motors, ii) lower energetic efficiency compared to unidirectional rotors, iii) lower controllability of the exerted force at low speeds, and iv) extra mechanical complexity and increased weight and thus energy consumption (in case of variable pitch propellers). Such propellers have the advantage that $\mathbf{u} \in \mathbb{R}^n$ in (41). Let us denote with U the set of admissible inputs: $U = \mathbb{R}_{\geq \mathbf{0}}^n$ for mono-directional thrusters and $U = \mathbb{R}^n$ for bidirectional propellers. Then let us define the set of attainable wrenches $W = \{\mathbf{w} \in \mathbb{R}^6 | \exists \mathbf{u} \in U \text{ s.t. } \mathbf{w} = \mathbf{G}\mathbf{u}\}$, the set of attainable total thrusts $\Phi = \{\mathbf{f} \in \mathbb{R}^3 | \exists \mathbf{u} \in U \text{ s.t. } \mathbf{w} = \mathbf{G}_1\mathbf{u}\}$, and the set of attainable total moments $\Theta = \{\boldsymbol{\tau} \in \mathbb{R}^3 | \exists \mathbf{u} \in U \text{ s.t. } \mathbf{w} = \mathbf{G}_2\mathbf{u}\}$.

The following additional classification of multirotor design holds.

Definition 3 A fully actuated multirotor design for which $\Theta = \mathbb{R}^3$ is a

- multi-directional total thrust platform if $\Phi \subsetneq \mathbb{R}^3$
- omnidirectional total thrust platform if $\Phi = \mathbb{R}^3$.

A multi-directional thrust platform can produce a total force (independent from total moment) in multiple (but not all) directions. As a consequence it can hover with multiple (but not all) orientations. An omnidirectional thrust platform can produce a total force (independent from total moment) in any direction and can hover with any orientation. Omnidirectional platforms can be oriented in any direction and can compensate/exert any force independently, thus allowing applications that are impossible with other platforms, including safe human interaction, 360° aerial photography, etc. In [225] and [41] two omnidirectional total thrust vehicles are proposed with 6 and 8 tilted bidirectional thrust rotors, respectively. As explained before, bidirectional thrusters have several drawbacks. The authors in [290] thoroughly investigate *if and how* it is instead possible to obtain omnidirectional thrust vehicles with fixed and uni-directional thrusters, a solution that overcome all the problems of the aforementioned solutions using bidirectional propellers. One of the main results is summarized in the following:

Proposition 1 ([290]) If $U = \mathbb{R}^n$ then a fully actuated multirotor is also omnidirectional. If $U = \mathbb{R}_{\geq \mathbf{0}}^n$ then a fully actuated multirotor is also omnidirectional if and only if

$$\text{null}(\mathbf{G}) \cap \mathbb{R}_{>0}^n \neq \emptyset, \quad (44)$$

otherwise, it is a multi-directional platform.

In other words, a fully-actuated platform with bidirectional propellers is also an omnidirectional total thrust platform. On the other side, a fully-actuated platform with mono-directional propellers is in general only a multi-directional total thrust platform. For such platforms, it is required that (44) holds too, which implies that $n \geq 7$. This latter is however only a necessary condition. In [290] the authors propose also a control allocation strategy and an algorithm for optimally design such platforms.

4 Coupling Between Total thrust and Total Moment

In the following we assume a fully-actuated rotational dynamics, i.e.,

$$\text{rank}(\mathbf{G}_2) = 3. \quad (45)$$

The input space \mathbb{R}^n can always be partitioned in the orthogonal subspaces $\text{im}(\mathbf{G}_2^\top)$ and $\text{im}(\mathbf{G}_2^\top)^\perp = \text{null}(\mathbf{G}_2)$, such that the vector \mathbf{u} can be rewritten as the sum of two terms, namely

$$\mathbf{u} = \mathbf{T}_2 \tilde{\mathbf{u}} = [\mathbf{A}_2 \ \mathbf{B}_2] \begin{bmatrix} \tilde{\mathbf{u}}_A \\ \tilde{\mathbf{u}}_B \end{bmatrix} = \mathbf{A}_2 \tilde{\mathbf{u}}_A + \mathbf{B}_2 \tilde{\mathbf{u}}_B, \quad (46)$$

where $\mathbf{T}_2 = [\mathbf{A}_2 \ \mathbf{B}_2] \in \mathbb{R}^{n \times n}$ is an orthogonal matrix such that $\text{im}(\mathbf{A}_2) = \text{im}(\mathbf{G}_2^\top)$ and $\text{im}(\mathbf{B}_2) = \text{null}(\mathbf{G}_2)$. Note that, because of (45), $\mathbf{A}_2 \in \mathbb{R}^{n \times 3}$ is full rank, i.e., $\text{rank}(\mathbf{A}_2) = 3$, while $\mathbf{B}_2 \in \mathbb{R}^{n \times n-3}$ has $\text{rank}(\mathbf{B}_2) = n - 3$. Given this partition, we have

$$\boldsymbol{\tau} = \mathbf{G}_2 \mathbf{T}_2 \tilde{\mathbf{u}} = \mathbf{G}_2 \mathbf{A}_2 \tilde{\mathbf{u}}_A, \quad (47)$$

$$\mathbf{f} = \mathbf{G}_1 \mathbf{T}_2 \tilde{\mathbf{u}} = \mathbf{G}_1 \mathbf{A}_2 \tilde{\mathbf{u}}_A + \mathbf{G}_1 \mathbf{B}_2 \tilde{\mathbf{u}}_B =: \mathbf{f}^A + \mathbf{f}^B. \quad (48)$$

The matrix $\mathbf{G}_2 \mathbf{A}_2$ in (47) is nonsingular thus any moment $\boldsymbol{\tau} \in \mathbb{R}^3$ can be virtually implemented by setting $\tilde{\mathbf{u}}_A = (\mathbf{G}_2 \mathbf{A}_2)^{-1} \boldsymbol{\tau}$ in conjunction with any $\tilde{\mathbf{u}}_B \in \mathbb{R}^{n-3}$.

The control force, which obviously belongs to $\mathfrak{F} := \text{im}(\mathbf{G}_1)$, is split in two components: $\mathbf{f} = \mathbf{f}^A + \mathbf{f}^B$. The component $\mathbf{f}^A = \mathbf{G}_1 \mathbf{A}_2 \tilde{\mathbf{u}}_A$ represents the ‘spurious’ force generated by the allocation of the input needed to obtain a non-zero control moment. This component belongs to the subspace $\mathfrak{F}_A := \text{im}(\mathbf{G}_1 \mathbf{A}_2) \subset \mathbb{R}^n$. The component $\mathbf{f}^B = \mathbf{G}_1 \mathbf{B}_2 \tilde{\mathbf{u}}_B$ instead represents a force that can be assigned independently from the control moment by allocating the input \mathbf{u} in $\text{im}(\mathbf{B}_2) = \text{null}(\mathbf{G}_2)$. This ‘free’ force component belongs to the subspace $\mathfrak{F}_B := \text{im}(\mathbf{G}_1 \mathbf{B}_2) \subset \mathbb{R}^n$ and it is obtained by assigning $\tilde{\mathbf{u}}_B$. Being \mathbf{T}_2 nonsingular, we have that $\mathfrak{F} = \mathfrak{F}_A + \mathfrak{F}_B$. It is instru-

	$\dim \mathfrak{F}_B = 0$	\exists decoupled direction		
		$\dim \mathfrak{F}_B = 1$	\exists decoupled plane	
			$\dim \mathfrak{F}_B = 2$	$\dim \mathfrak{F}_B = 3$
$\mathfrak{F}_B \subsetneq \mathfrak{F}$	FC	PC and SD1	PC and SD2	N/A
$\mathfrak{F}_B = \mathfrak{F}$	N/A	UC and SD1	UC and SD2	D3 (UC)
	$(\dim \mathfrak{F} \geq 1)$	$(\dim \mathfrak{F} \geq 1)$	$(\Rightarrow \dim \mathfrak{F} \geq 2)$	$(\Rightarrow \dim \mathfrak{F} = 3)$

Table 9 A table recalling the fundamental properties of the actuation of a GOFP.

mental to recall that $1 \leq \dim \mathfrak{F} \leq 3$ because $\text{rank}(\mathbf{G}_1) \geq 1$, and that $\mathfrak{F}_B \subseteq \mathfrak{F}$, thus $\dim \mathfrak{F} \geq \dim \mathfrak{F}_B$.

The dimension of \mathfrak{F}_B and its relation with \mathfrak{F} sheds light upon the platform actuation capabilities. The following two sets of definitions are devoted to this purpose.

Definition 1. A platform with generically-oriented fixed propellers (GOFP) is

- *fully coupled* (FC) if $\dim \mathfrak{F}_B = 0$ (i.e., if $\mathbf{G}_1 \mathbf{B}_2 = \mathbf{0}$)
- *partially coupled* (PC) if $\dim \mathfrak{F}_B \in \{1, 2\}$ and $\mathfrak{F}_B \subsetneq \mathfrak{F}$
- *un-coupled* (UC), or fully-decoupled, if $\mathfrak{F}_B = \mathfrak{F}$ (or, equivalently, $\mathfrak{F}_A \subseteq \mathfrak{F}_B$)

In a fully coupled GOFP the control force depends completely upon the implemented control moment, in fact $\mathbf{f}^B = \mathbf{0}$ and thus $\mathbf{f} = \mathbf{f}^A$. In a partially coupled GOFP the projection of the control force onto \mathfrak{F}_B can be chosen freely while the projection onto $\mathfrak{F}_B^\perp \cap \mathfrak{F}$ depends completely upon the implemented control moment. Finally in a un-coupled (equivalently, fully decoupled) GOFP no projection of the control force depends on the control moment, i.e., the control force can be freely assigned in the whole space \mathfrak{F} . Notice that the full decoupling does not imply necessarily that the control force can be chosen in the whole \mathbb{R}^3 , unless it holds also $\mathfrak{F} = \mathbb{R}^3$.

The second important classification is provided in the following definition.

Definition 2. A GOFP

- has a *decoupled direction* (D1) if $\dim \mathfrak{F}_B \geq 1$
- has a *decoupled plane* (D2) if $\dim \mathfrak{F}_B \geq 2$
- is *fully actuated* (D3) if $\dim \mathfrak{F}_B = 3$.

Combining the previous definitions we say that a GOFP

- has a *single decoupled direction* (SD1) if $\dim \mathfrak{F}_B = 1$
- has a *single decoupled plane* (SD2) if $\dim \mathfrak{F}_B = 2$.

If a GOFP has a decoupled direction then there exists at least a direction along which the projection of the control force can be chosen freely from the control moment. If a GOFP has a decoupled plane then there exists at least a plane over which the projection of the control force can be chosen freely from the control moment. If a GOFP is fully actuated then the control force can be chosen in all \mathbb{R}^3 freely from the control moment.

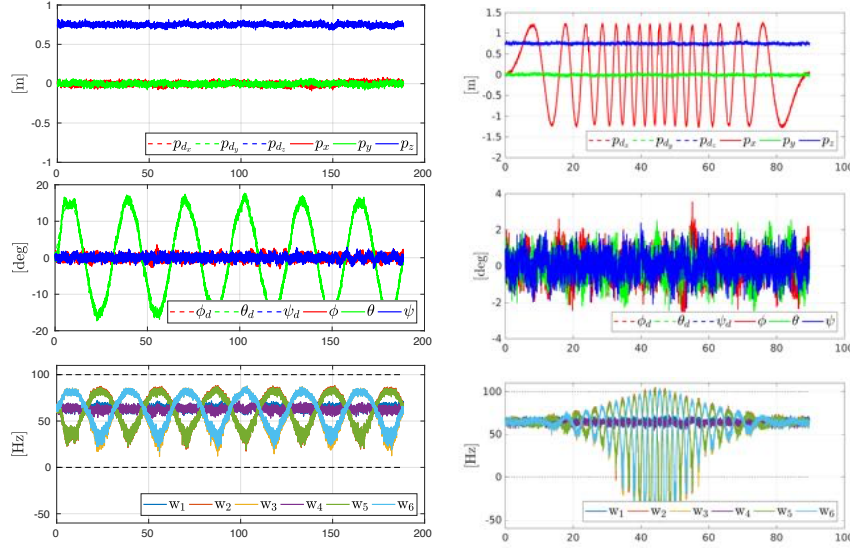


Fig. 42 Two realistic simulations (noise and model uncertainty included) of the TiltHex in Fig. 41 controlled via the inverse dynamics approach. Left column: the TiltHex by LAAS-CNRS follows a trajectory in which the position is constant and the roll varies sinusoidally. Right column: the TiltHex by LAAS-CNRS follows a trajectory in which the orientation is constant and the position follows a chirp signal. Both trajectories are unfeasible for the standard hexarotor in Fig. 40.

In terms of relations between the above definitions, we note that: D3 implies UC, while the converse is not true; D3 implies D2; D2 implies D1. Finally, D1 (and thus D2) can coexist with PC or UC but not with FC. Note that in the state-of-the-art multirotor controllers it is implicitly assumed that the GOFD is fully decoupled and there exists a decoupled direction oriented along its \mathbf{z}_B axis. Nevertheless, in the controller proposed in [184] the decoupled direction can be any and the GOFD can be also partially coupled. Table 9 yields a comprehensive view of all aforementioned definitions and relations. For more examples and insights about GOFD's, as well as for a study of their ability to statically hover after propeller losses we refer the reader to [185].

5 Inverse Dynamics Control for Fully-Actuated Vehicles

Plugging (41) in (40), we obtain

$$\mathbf{a} = \mathbf{M}_R^{-1}(\mathbf{b}_R + \mathbf{G}_R \mathbf{G} \mathbf{u}). \quad (49)$$

According to Definition 2 the allocation matrix \mathbf{G} is full-rank for a fully-actuated aerial vehicle, therefore one can choose the following inverse dynamics control law:

$$\mathbf{u} = \mathbf{G}^\dagger \mathbf{G}_R^{-1} (\mathbf{M}_R \mathbf{v} - \mathbf{b}_R), \quad (50)$$

which brings the system (49) in the following linear and decoupled form:

$$\mathbf{a} = \mathbf{v}, \quad (51)$$

where \mathbf{v} is a six-dimensional virtual input which can be assigned at will in order to independently steer the six degrees of freedom of the platform along any trajectory, and to ensure robustness of the control scheme. Such classical nonlinear control scheme has been proposed in [237] for fully-actuated aerial vehicles. Figure 42 shows two realistic simulations of the TiltHex executing two trajectories that are unfeasible for a standard collinear hexarotor using the inverse dynamics approach.

The main limitation of the inverse dynamics approach is that, as it is, it does not take into account the input saturations. This fact can easily destabilize the platform when unfeasible inputs are needed in order to perfectly track a given trajectory. This fact can be seen in the simulation of Fig. 43, where the same reference trajectory of Fig. 42-right is used but for the introduction of input limits. In order to overcome this drawback, a more clever controller should be used, as the one that is presented in Chapter 3.6.

6 Conclusions

In this Chapter we have briefly presented the multi-directional thrust aerial platforms emerging from a generic orientation of the propeller angular velocity vectors, i.e., GOFP platforms. We have provided the fundamental definitions of multi-directionality and omnidirectionality and a classification based on the decoupling between the total moment and the total force. In Chapter 3.6 of this book it will be shown how such platforms can be effectively used for aerial physical interaction. One of the most interesting properties of GOFP platforms is their ability, under certain condition, to gain the property to statically hover after propeller losses. Due to the limited amount of space such topic has not been addressed in this chapter, however the reader can find an extensive description of such property in [185].

7 Acknowledgements

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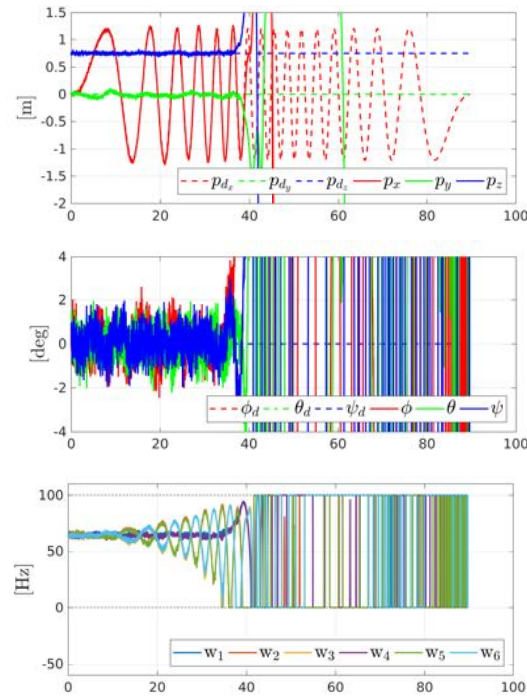


Fig. 43 A realistic simulation (noise and model uncertainty included) of the TiltHex by LAAS-CNRS in Fig. 41 with the addition of input saturation. While tracking the same trajectory of Fig. 42-right the sole inverse dynamics approach makes the system unstable. A controller that overcomes such pitfall is presented in Chapter 3.6.

Chapter 3.6

Interaction Control of Platforms with Multi-directional Total Thrust

Antonio Franchi

Abstract The chapter introduces an interaction control framework for multi-directional total thrust platforms. When also fully-actuated, such platforms can modify the total wrench in body frame in any direction. Therefore, they do not suffer from the under-actuation of standard collinear multirotors, and are best suited for dexterous tasks, physical interaction, and for carrying aerial manipulators. The chapter describes in order: a full-pose controller which takes into account the lateral limits of the total force; an algorithm for estimating the contact wrench; and an admittance-shaping framework for physical interaction. Experimental results with the TiltHex platform are also shown to validate the proposed methods.

1 Introduction

Standard multi-rotors (quadrotors, hexarotors, etc.) have collinear propellers generating a total force that is aligned to one direction in body frame, which makes them under-actuated systems. In order to follow a generic position trajectory the total force direction in world frame is changed by rotating the whole vehicle. Maneuvers in which rotation and translation are completely independent are precluded to such platforms, which constitutes a serious problem in the case that, e.g., the platform has to move through a hostile and cluttered ambient or resist a wind gust while keeping a desired attitude. Such an underactuation even deteriorates the potentiality to interact with the environment because it makes impossible to rapidly exert forces in an arbitrarily-chosen direction in the space while keeping a pre-specified orientation.

As described in Chapter 2.3, the main solution to overcome the aforementioned issues has been to mount rotors in a tilted way such that the thrusts of the propellers are not collinear anymore. In this way, the direction of the total force can be changed by selecting the intensity of the force produced by each propeller, without the need of reorienting the whole vehicle. If the propellers are at least six, and the tilting directions do not generate a singular configuration, then direction and intensity of

both the instantaneous total control moment and total control force are, in principle, controllable at will.

However, in order to minimize the waste of energy caused by the appearance of internal forces, the maximum component of the total thrust along the lateral direction is typically kept (by design) much lower than the maximum allowed component along the vertical one. We call these kind of platforms aerial vehicles with *laterally-bounded force* (LBF): they are characterized by a *principal* direction of thrust along which most of the thrust can be exerted. A certain amount of thrust (typically smaller) can be exerted along any non-principal (lateral) directions. This model includes: *i*) the standard underactuated multi-rotor vehicle where thrust is possible *only* along the principal direction, and *ii*) the isotropically fully-actuated platforms where a large amount of total thrust in the lateral directions is applicable [225, 41, 290].

This Chapter is structured as follows. Section 2 presents the generic model of a LBF platforms. Section 3 presents a theoretically grounded 6D tracking control algorithm for LBF platforms is introduced in Sec. 3. Section 4 shows the design of a full architecture to enable 6D interaction control using an admittance control scheme, built around the tracker described in Sec. 3. Results of experiments for the contact-free tracking and interaction control in contact are shown in Sec. 5.1 and 5.2. Finally we conclude the paper and give an outline of further possible extensions in Sec. 6.

2 Model of Laterally-Bounded Force Platforms

In this Section we propose the concept of Laterally-Bounded Force (LBF) Platform, which represents a good approximation of any multi-directional total thrust platform. An LBF platform is a rigid body to which gravity and control generalized forces are applied. All the main symbols are summarized in Table 15 and Fig. 112. Some of the symbols have been already defined at the beginning of the book and are repeated here in order to ease the reading. For the sake of reducing the heaviness of notation, the use of the subscript \star_b is implied in the position, rotation matrix and angular velocity notations. We consider here flying platforms which do not carry any manipulator. Therefore, without risk of confusion, within this Chapter we can denote with \mathbf{q} the configuration (\mathbf{p}, \mathbf{R}) of the vehicle rather than the one of the manipulator.

The vehicle orientation kinematics is described by $\dot{\mathbf{R}} = \mathbf{R}[\boldsymbol{\omega}]$, where $[\boldsymbol{\omega}]_{\times} \in so(3)$ is the skew symmetric matrix associated to $\boldsymbol{\omega}$. The control inputs $\mathbf{u}_1 = [u_1 \ u_2 \ u_3]^T \in \mathbb{R}^3$ and $\mathbf{u}_2 = [u_4 \ u_5 \ u_6]^T \in \mathbb{R}^3$ are the input force and moment applied to the vehicle expressed in \mathcal{F}_B . The following constraint applies

$$[u_1 \ u_2]^T \in \mathcal{U}_{xy} \subset \mathbb{R}^2, \quad (142)$$

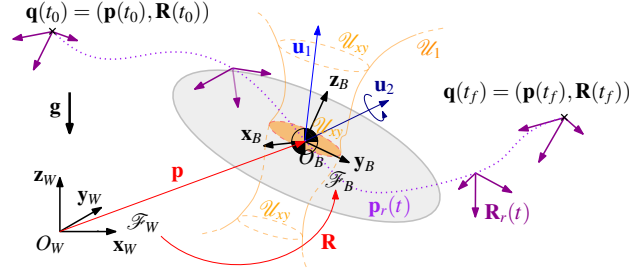


Fig. 112 A drawing illustrating the main quantities of an LBF Aerial Vehicle, the main frames involved, the laterally bounded input sets and the full-pose 6D reference trajectory.

Table 15 Symbols use to define the LBF platform model.

Definition	Symbol
World Inertial Frame: $\mathcal{F}_W = O_W, \{\mathbf{x}_W, \mathbf{y}_W, \mathbf{z}_W\}$	\mathcal{F}_W
Attached Body Frame: $\mathcal{F}_B = O_B, \{\mathbf{x}_B, \mathbf{y}_B, \mathbf{z}_B\}$	\mathcal{F}_B
Position of O_B in \mathcal{F}_W , coincident with the Center of Mass (CoM)	$\mathbf{p}_b = \mathbf{p}$
Rotation matrix mapping coordinates in \mathcal{F}_B to coordinates in \mathcal{F}_W	$\mathbf{R}_b = \mathbf{R}$
Configuration of the vehicle	$\mathbf{q} = (\mathbf{p}, \mathbf{R})$
Angular velocity of \mathcal{F}_B w.r.t \mathcal{F}_W expr. in \mathcal{F}_B	$\boldsymbol{\omega}_b = \boldsymbol{\omega}$
Vehicle's Inertia matrix w.r.t to O_B expressed in \mathcal{F}_B	\mathbf{I}_b
Control force applied at the CoM expressed in \mathcal{F}_B	\mathbf{u}_1
Control moment applied at the CoM expressed in \mathcal{F}_B	\mathbf{u}_2
Feasible set of the control force \mathbf{u}_1	\mathcal{U}_1
Feasible set of the projection of \mathbf{u}_1 on the xy plane in \mathcal{F}_B	\mathcal{U}_{xy}
i -th vector of the canonical basis of \mathbb{R}_3 with $i = 1, 2, 3$	\mathbf{e}_i

where the *laterally bounding* set \mathcal{U}_{xy} is a set that contains the origin. We define $\mathcal{U}_1 = \{\mathbf{u}_1 \in \mathbb{R}^3 \mid [u_1 \ u_2]^T \in \mathcal{U}_{xy}, u_3 \geq 0\}$. Note that \mathcal{U}_{xy} can be constant or even be changing depending of u_3 , as shown in Figure 112. The dynamics of the aerial platform is then

$$m\ddot{\mathbf{p}} = -mg\mathbf{e}_3 + \mathbf{R}\mathbf{u}_1, \quad \mathbf{I}_b\dot{\boldsymbol{\omega}} = -\boldsymbol{\omega} \times \mathbf{I}_b\boldsymbol{\omega} + \mathbf{u}_2. \quad (143)$$

Some particularly relevant cases of LBF follow.

Case 1 (Underactuated aerial vehicle) When $\mathcal{U}_{xy} = \{\mathbf{0}\}$ the total force is always oriented as $\mathbf{R}\mathbf{e}_3$ and model (143) becomes the standard underactuated quadrotor model.

Case 2: (Conic LBF) When $\mathcal{U}_{xy} = \{[u_1 \ u_2]^T \in \mathbb{R}^2 \mid u_1^2 + u_2^2 \leq (\tan \alpha)^2 u_3^2\}$, model (143) approximates the case of hexarotors with tilted propellers [237, 254], for which the set of allowable \mathcal{U}_1 forces has the pseudo-inverted-pyramidal shape. The quantity α is a parameter that represents the tilting angle of the propellers (hexarotor).

Case 3: (Cylindric LBF) When

$$\mathcal{U}_{xy} = \{[u_1 \ u_2]^T \in \mathbb{R}^2 \mid u_1^2 + u_2^2 \leq r_{xy}^2\}, \quad (144)$$

Table 16 Symbols related to the LBF platform tracking control.

Definition	Symbol
Reference position for \mathbf{p} at time t	$\mathbf{p}_r(t)$
Reference rotation matrix for \mathbf{R} at time t	$\mathbf{R}_r(t)$
Reference control force to be applied at O_B	$\mathbf{f}_r(t)$ at time t
Set of orientations in $SO(3)$ that allow the application of $\mathbf{f}_r(t)$	$\mathcal{R}(\mathbf{f}_r)$
Subset of $\mathcal{R}(\mathbf{f}_r)$ that minimizes a certain cost w.r.t. \mathbf{R}_r	$\overline{\mathcal{R}}(\mathbf{f}_r, \mathbf{R}_r)$
Desired rotation matrix in $\overline{\mathcal{R}}(\mathbf{f}_r, \mathbf{R}_r)$	\mathbf{R}_d

model (143) approximates the case of an multirotor with n_m main propellers pointing up and n_s secondary less powerful propellers tilted 90 degrees w.r.t. the main ones, like the one presented in [244], for which the set of allowable \mathcal{U}_1 forces can be approximated by a pseudo-cylindric shape

3 Full-pose Tracking Control for LBF Platforms

An underactuated LBF platform (the Case 1 depicted before) is not able to track a generic full-pose trajectory, i.e., with independent position \mathbf{p} and orientation \mathbf{R} in $SE(3)$. The rotation about any axis that is orthogonal to \mathbf{z}_B must follow the evolution over time of \mathbf{p} , according to the well-known differential flatness property [188, 79]. Only the tracking of a 4D-pose trajectory (i.e., position \mathbf{p} plus the rotation about \mathbf{z}_B) is possible. On the contrary, a ‘fully-actuated’ LBF platform can exert some force in the lateral direction thus allowing the tracking of a generic full-pose (6D) trajectory. However, due to the bounded thrust along the lateral directions, it is not possible to track *any* full-pose trajectory. The larger the bounds the higher the ability of the platform to track any trajectory, the lower the bounds the more the platform resembles an underactuated multi-rotor and thus it becomes almost unable to track a full-pose trajectory but only a 4D-pose one.

The most straightforward approach to control fully-actuated platforms is the inverse dynamics approach. First, a *control wrench* is computed in order to track the desired trajectory by cancelling the nonlinear dynamical effects and trying to zero the position and orientation errors. Then the thrust inputs for each propeller are computed from the control wrench by simply inverting the control allocation matrix. This method has been first proposed in [237] and then used also in [41] (with pseudo-inversion, in place of inversion, to allocate the input redundancy) and in [225]. The limitation of this control approach is to neglect input saturation, which may easily lead to an unstable behavior if, e.g., the full-pose trajectory to be followed is not input-feasible.

In this Section we briefly present a geometric tracking controller for time-varying references, introduced in [87], that is instead very general and applicable to any LBF vehicle, thus also taking into account the bounds on the lateral control force. The

method is not prone to local orientation representation singularities since it is natively designed in $SE(3)$. The proposed controller ensures, in nominal conditions, the tracking of a full-6D pose reference trajectory (position plus orientation). If the reference orientation and the force needed to track the position trajectory do not comply with the platform constraints, the proposed strategy gives priority to the tracking of the positional part while also tracking the feasible orientation that is the closest to the reference one. This choice is supported by, e.g., the fact that in typical applications a wrong position tracking is more likely to lead to an obstacle crash than a non-perfect orientation tracking. The proposed method also suitable to control vectored-thrust vehicles that can transit from an under-actuated to a fully-actuated configuration while flying – as, e.g., the one presented in [254]. The proposed method is in this sense ‘universal’, since it does *not* need any switching between two different controllers for each configuration.

Let be given a full-pose trajectory $\mathbf{q}_r(t) = (\mathbf{p}_r(t), \mathbf{R}_r(t)) : [t_0, t_f] \rightarrow SE(3)$, where $\mathbf{p}_r(t) \in \mathbb{R}^3$ is the reference position trajectory and $\mathbf{R}_r(t) \in SO(3)$ is the reference attitude trajectory (see Table 16 for a recap of the symbols used in this Section). Inverting (143), the nominal inputs to track $\mathbf{q}_r(t)$ are obtained as $\mathbf{u}_1^r = \mathbf{R}_r^T (m\mathbf{g}\mathbf{e}_3 + m\dot{\mathbf{p}}_r)$ and $\mathbf{u}_2^r = \boldsymbol{\omega}_r \times \mathbf{I}_b \boldsymbol{\omega}_r + \mathbf{I}_b \dot{\boldsymbol{\omega}}_r$, where $\boldsymbol{\omega}_r$ is defined by $[\boldsymbol{\omega}_r]_\times = \mathbf{R}_r^T \dot{\mathbf{R}}_r$.

Definition 4 $\mathbf{q}_r(t)$ is feasible if $\mathbf{u}_1^r(t) \in \mathcal{U}_1 \forall t \in [t_0, t_f]$.

Exact full-pose (6D) tracking is possible only if $\mathbf{q}_r(t)$ is feasible. However in real world it is not granted that $\mathbf{q}_r(t)$ will be such for the particular LBF platform in use. For this reason, we propose a controller that works (in the sense that the tracking of $\mathbf{p}_r(t)$ is still guaranteed and no singularity appears) even if $\mathbf{q}_r(t)$ is not feasible. Consider the position error $\mathbf{e}_p = \mathbf{p} - \mathbf{p}_r$, velocity error $\mathbf{e}_v = \dot{\mathbf{p}} - \dot{\mathbf{p}}_r$, and two positive definite gain matrices \mathbf{K}_p and \mathbf{K}_v . Then consider $\mathbf{f}_r = m\ddot{\mathbf{p}}_r + m\mathbf{g}\mathbf{e}_3 - \mathbf{K}_p \mathbf{e}_p - \mathbf{K}_v \mathbf{e}_v$, representing the reference total control force that ideally one would like to apply to the aerial vehicle CoM if the system would be completely fully actuated, i.e., if $\mathcal{U}_1 = \mathbb{R}^3$.

The set of orientations that allow to apply \mathbf{f}_r to the CoM of the LBF aerial vehicle is defined as $\mathcal{R}(\mathbf{f}_r) = \{\mathbf{R} \in SO(3) \mid \mathbf{R}^T \mathbf{f}_r \in \mathcal{U}_1\}$. For an underactuated collinear multi-rotor system the set $\mathcal{R}(\mathbf{f}_r)$ is formed by any \mathbf{R} such that $\mathbf{R}\mathbf{e}_3$ and \mathbf{f}_r are parallel, i.e., $\mathbf{R}\mathbf{e}_3 \times \mathbf{f}_r = 0$. For a generic LBF aerial vehicle the set $\mathcal{R}(\mathbf{f}_r)$ may contain also \mathbf{R} 's for which $\mathbf{R}\mathbf{e}_3 \times \mathbf{f}_r \neq 0$. It is possible to show that (see [87] for its proof):

Proposition 2 The set $\mathcal{R}(\mathbf{f}_r)$ is always nonempty $\forall \mathbf{f}_r \in \mathbb{R}^3$.

The proposed controller exploits a cascaded structure³ by choosing, at each time t , a desired orientation $\mathbf{R}_d \in SO(3)$ that belongs to $\mathcal{R}(\mathbf{f}_r)$ and also minimizes a given cost function w.r.t. \mathbf{R}_r . Then one can use the fully actuated rotational dynamics to track \mathbf{R}_d and, in turn, track the reference position \mathbf{p}_r . If \mathbf{q}_r is feasible then \mathbf{R}_d will exponentially converge to \mathbf{R}_r . Otherwise, only the best feasible orientation will be obtained. Therefore the controller implicitly prioritizes the position trajectory tracking against the orientation one.

³ Notice that even if a cascaded structure is used, there is no time-scale separation assumption in the proposed controller.

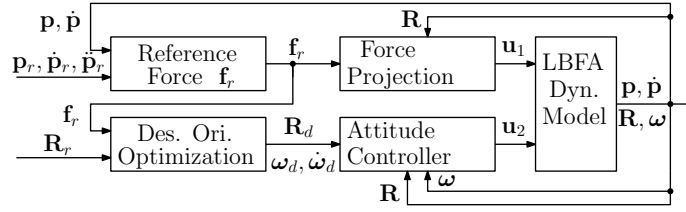


Fig. 113 Block diagram of the 6D tracker for LBF platforms.

Define $\overline{\mathcal{H}}(\mathbf{f}_r, \mathbf{R}_r) \subset \mathcal{H}(\mathbf{f}_r)$ as the set of rotation matrices that solve

$$\min_{\mathbf{R}' \in \mathcal{H}(\mathbf{f}_r)} J(\mathbf{R}_r, \mathbf{R}'),$$

where $J : SO(3) \times SO(3) \rightarrow \mathbb{R}_{\geq 0}$ is an arbitrarily chosen cost function that represents the degree of similarity between \mathbf{R}_r and \mathbf{R}' one is interested in. The elements in $\overline{\mathcal{H}}(\mathbf{f}_r, \mathbf{R}_r)$ represent orientations of the LBF that allow to apply \mathbf{f}_r and minimize the function J w.r.t. \mathbf{R}_r .

Consider that, at each time t a desired orientation $\mathbf{R}_d \in \overline{\mathcal{H}}(\mathbf{f}_r, \mathbf{R}_r)$ is chosen. Furthermore, whenever $\mathbf{R}_r \in \overline{\mathcal{H}}(\mathbf{f}_r, \mathbf{R}_r)$ then \mathbf{R}_d must be chosen equal to \mathbf{R}_r . Then define the rotation error $\mathbf{e}_R = \frac{1}{2}(\mathbf{R}_d^T \mathbf{R} - \mathbf{R}^T \mathbf{R}_d)^\vee$, and the angular velocity error $\mathbf{e}_\omega = \boldsymbol{\omega} - \mathbf{R}^T \mathbf{R}_d \boldsymbol{\omega}_d$ where \bullet^\vee is the inverse map of $[\star]_\times$, and $\boldsymbol{\omega}_d$ is the angular velocity associated to \mathbf{R}_d . Consider then the following control law

$$\mathbf{u}_1 = \text{sat}_{\mathcal{U}_{xy}}((\mathbf{f}_r^T \mathbf{R} \mathbf{e}_1) \mathbf{e}_1 + (\mathbf{f}_r^T \mathbf{R} \mathbf{e}_2) \mathbf{e}_2) + (\mathbf{f}_r^T \mathbf{R} \mathbf{e}_3) \mathbf{e}_3 \quad (145)$$

$$\mathbf{u}_2 = \boldsymbol{\omega} \times \mathbf{I}_b \boldsymbol{\omega} - \mathbf{K}_R \mathbf{e}_R - \mathbf{K}_\omega \mathbf{e}_\omega - \mathbf{I}_b([\boldsymbol{\omega}]_\times \mathbf{R}^T \mathbf{R}_d \boldsymbol{\omega}_d - \mathbf{R}^T \mathbf{R}_d \dot{\boldsymbol{\omega}}_d) \quad (146)$$

where $\text{sat}_{\mathcal{U}_{xy}}(\mathbf{x})$ is a vector in \mathcal{U}_{xy} with the same direction of \mathbf{x} , that minimizes the distance from \mathbf{x} . $\mathbf{K}_R = k_R \mathbf{I}$ and $\mathbf{K}_\omega = k_\omega \mathbf{I}$ are the gain matrices with $k_R > 0$ and $k_\omega > 0$.

In order to state the convergence properties of the proposed controller let us consider the following error function between two rotation matrixes \mathbf{R}_1 and \mathbf{R}_2 to be $d(\mathbf{R}_1, \mathbf{R}_2) = \frac{1}{2} \text{tr}(\mathbf{I} - \mathbf{R}_2^T \mathbf{R}_1)$. the following result holds (see [87] for the proof):

Theorem 1 Assume that $\mathbf{R}_d(t) \in \mathcal{H}(\mathbf{f}_r(t))$ for any t and that $\boldsymbol{\omega}_d(t)$ and $\dot{\boldsymbol{\omega}}_d(t)$ are well defined for any t . Consider the control \mathbf{u}_1 and \mathbf{u}_2 defined at (145) and (146).

Assume that the initial condition satisfies

$$d(\mathbf{R}(0), \mathbf{R}_d(0)) < 2, \quad (147)$$

$$\|\mathbf{e}_\omega(0)\|^2 < \frac{2}{\lambda_{\min}(\mathbf{I}_b)} k_R (1 - d(\mathbf{R}(0), \mathbf{R}_d(0))) \quad (148)$$

Then, the zero equilibrium of the tracking errors \mathbf{e}_R , \mathbf{e}_ω , \mathbf{e}_p and \mathbf{e}_v is exponentially stable. The region of attraction is characterized by (147) and (148).

A block diagram that shows the main subsystems of the proposed control architecture is provided in Fig. 113.

Theorem 1 ensures, under mild conditions, the exponential stability of \mathbf{e}_p , \mathbf{e}_v , \mathbf{e}_R , and \mathbf{e}_ω . Notice that this results holds regardless of the feasibility of \mathbf{q}_r . If \mathbf{q}_r is also feasible then exponential tracking of \mathbf{q}_r by \mathbf{q} is also guaranteed. In order to formally state this fact let us define the errors $\mathbf{e}_{R_r} = \frac{1}{2}(\mathbf{R}_r^T \mathbf{R}_d - \mathbf{R}_d^T \mathbf{R}_r)^\vee$, and $\mathbf{e}_{\omega_r} = \boldsymbol{\omega}_d - \mathbf{R}_d \mathbf{R}_r^T \boldsymbol{\omega}_r$.

In next result we characterize the convergence of the above errors to zero provided that the reference trajectory $\mathbf{q}_r(t)$ is *feasible* and satisfies the additional property that \mathbf{u}_1^r is *sufficiently inside* \mathcal{U}_1 , meaning that there exists a time instant \bar{t} and a positive number ε such that the distance of \mathbf{u}_1^r from the boundary of \mathcal{U}_1 is greater than $\varepsilon > 0$ for all $t > \bar{t}$, i.e.,

$$\text{dist}(\mathbf{u}_1^r(t), \partial \mathcal{U}_1) > \varepsilon, \quad \forall t > \bar{t}. \quad (149)$$

Corollary 1 *Assume $\mathbf{q}_r(t)$ is a feasible trajectory and that it satisfies the additional property in (149). Assume that $\mathbf{R}_d(t) \in \mathcal{R}(\mathbf{f}_r(t))$ for any t and that $\boldsymbol{\omega}_d(t)$ and $\dot{\boldsymbol{\omega}}_d(t)$ are well defined for any t . Consider the control \mathbf{u}_1 and \mathbf{u}_2 defined at (145) and (146). Assume that the initial condition satisfies (147) and (148). Then the zero equilibrium of the tracking errors \mathbf{e}_R , \mathbf{e}_ω , \mathbf{e}_p and \mathbf{e}_v is exponentially stable and there exists a time instant $\bar{t} \geq t_0$ such that $\mathbf{e}_{R_r}(t) = \mathbf{e}_{\omega_r}(t) = 0$ for all $t > \bar{t}$. The region of attraction is characterized by (147) and (148).*

The proposed controller (in particular the attitude controller (146)) relies on the availability of $\boldsymbol{\omega}_d$, and $\dot{\boldsymbol{\omega}}_d$. These quantities depend in turn on \mathbf{R}_d which is the output of an optimization algorithm executed at each control step. In order for $\boldsymbol{\omega}_d$ and $\dot{\boldsymbol{\omega}}_d$ to be well defined and available the optimization must ensure a sufficient smoothness of \mathbf{R}_d . This could be enforced by adding, e.g., a regularization term in the cost function J . If in the real case at hand this is not possible (or not implementable), then at each time instant in which \mathbf{R}_d is not smooth the attitude controller will undergo a new transient phase. In practice, see the experiments in Sec. 5.1, we have experimentally ascertain that the presence of a few isolated non-smooth instants does not constitute at all a real problem for the stability of the implementation and that regularization is actually not needed for practical stabilization.

The proposed control method is kept on purpose general regarding two main features: the choice of \mathcal{U}_{xy} in (142) and the choice of the cost function J . The former allows the method to be used for a large set of aerial vehicles with different actuation capabilities. The latter allows the engineer to customize the definition of similarity between two orientations in order to comply with the particular task at hand. It is An illustration of how these two general features are particularized for a specific meaningful case is provided in [87].

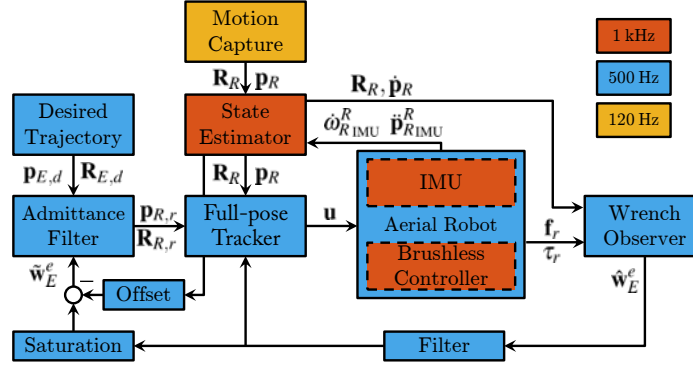


Fig. 114 Signal block diagram of the control framework. The runtime frequency is highlighted. For clarity higher derivatives of the signals have been omitted. The cascaded structure of the pose controller has been omitted as well.

4 Control of the Interaction

In this Section we illustrate how the 6D full-pose tracking controller introduced in Sec. 3 can be integrated in a larger architecture in order to allow the control of the physical interaction with multi-directional thrust platforms.

The control framework is based on an outer loop admittance control and an inner loop full-pose tracking controller (see Fig. 114). The state of the aerial robot is estimated by a Unscented Kalman Filter (UKF) that fuses the Inertial Measurement Unit (IMU) acceleration and angular velocity measurements with the position and orientation from a pose sensor (e.g., a motion capture system or an onboard camera using a Perspective-n-Point (PnP) algorithm). The interaction torques and forces are estimated by a wrench observer. We will now introduce all single components, except for the full-pose tracking controller, already introduced in Sec. 3.

4.1 Contact Wrench Estimation

In order to properly handle physical interaction of the aerial robot with the external environment, the knowledge of the contact interaction wrench between the tool tip and the environment, $\mathbf{w}_E = [\mathbf{f}_E^T \ \tau_E^T]^T \in \mathbb{R}^6$ is essential. To this aim, a force/torque sensor could be mounted on the robot's tool-tip, which is usually capable to provide a reliable measure, but this solution increases both the cost and the weight of the robot. In the aerial robotics field, a more viable solution is the adoption of a wrench estimator, that can provide a sufficiently accurate estimation, denoted as $\hat{\mathbf{w}}_E = [\hat{\mathbf{f}}_E^T \ \hat{\tau}_E^T]^T \in \mathbb{R}^6$, in the presence of accurate measurements of position, velocities and, if available, accelerations.

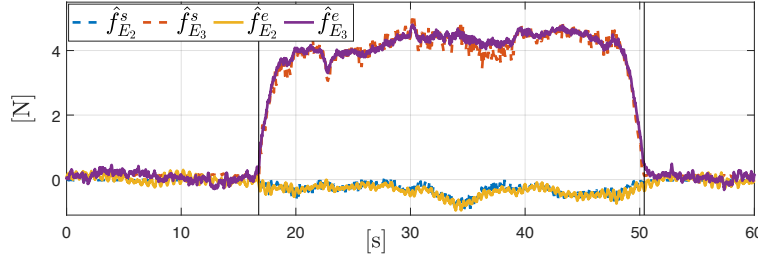


Fig. 115 Estimated versus measured contact force. The ground truth is measured with an ATI45 force torque transducer. The aerial robot pushed the sensor vertically. Both signals have been filtered with a low-pass with a 6 Hz cut-off frequency. For reasons of clarity only the second and third components are presented as the first and second component are overlapping each other a lot.

The external wrench on the robot, $\mathbf{w}_R = [\mathbf{f}_R^T, \mathbf{R}_R \boldsymbol{\tau}_R^T]^T$, can be viewed as the effect on the robot CoM of the wrench \mathbf{w}_E exerted by the environment on the tool tip, namely

$$\mathbf{w}_R = \mathbf{H}_E^T(\mathbf{R}_R) \mathbf{w}_E, \quad \mathbf{H}_E(\mathbf{R}_R) = \begin{bmatrix} \mathbf{I}_3 & -[\mathbf{R}_R \mathbf{p}_E^R]_\times \\ \mathbf{O}_3 & \mathbf{I}_3 \end{bmatrix}. \quad (150)$$

If the sensor equipment provides accurate enough measurements of the platform position and velocities, both angular and linear, while only the linear acceleration, provided by the IMU, can be reasonably used in a wrench observer. Thus we propose the hybrid approach already proposed in [293]. More in detail, the acceleration based observer proposed by [316] is adopted in order to estimate the external interaction forces on the robot CoM, \mathbf{f}_R , while the external torques, $\boldsymbol{\tau}_R^R$ are obtained by exploiting a momentum-based observer ([63]).

Estimation of contact forces. The following disturbance observer, firstly proposed for aerial robots in [316], is adopted for estimating the contact forces

$$\dot{\hat{\mathbf{f}}}_R = \mathbf{L}(\mathbf{f}_R - \hat{\mathbf{f}}_R) = -\mathbf{L}\hat{\mathbf{f}}_R - \mathbf{L}(m\ddot{\mathbf{p}}_R + mg\mathbf{e}_3 - \mathbf{R}_R \mathbf{F}_1 \mathbf{u}), \quad (151)$$

where $\mathbf{L} \in \mathbb{R}^{3 \times 3}$ is a gain matrix to be designed. By defining the observer error as $\mathbf{e}_f = \mathbf{f}_R - \hat{\mathbf{f}}_R$, the error dynamics, in the presence of a constant or slowly varying external force is given by [316]

$$\dot{\mathbf{e}}_f + \mathbf{L}\mathbf{e}_f = \mathbf{0}. \quad (152)$$

Thus, the error dynamics is asymptotically convergent to the origin for any positive definite matrix \mathbf{L} .

Estimation of contact moments. With reference to the system dynamics (143), the angular momentum $\mathbf{q}^R \in \mathbb{R}^3$ in frame \mathcal{F}_R can be computed as

$$\mathbf{q}^R = \mathbf{J}\boldsymbol{\omega}_R. \quad (153)$$

The time-derivative of (153) can be expressed as

$$\dot{\mathbf{q}}^R = \mathbf{J}\dot{\boldsymbol{\omega}}_R = -\boldsymbol{\omega}_R \times \mathbf{J}\boldsymbol{\omega}_R + \mathbf{F}_2\mathbf{u} + \boldsymbol{\tau}_R^R. \quad (154)$$

By exploiting (154), the estimate $\hat{\boldsymbol{\tau}}_R$ can be seen as the residual vector

$$\hat{\boldsymbol{\tau}}_R^R = \mathbf{K}_I \left[(\mathbf{q}^R(t) - \mathbf{q}^R(t_0)) + \int_{t_0}^t (\boldsymbol{\omega}_R \times \mathbf{J}\boldsymbol{\omega}_R - \mathbf{F}_2\mathbf{u} - \hat{\boldsymbol{\tau}}_R^R) d\tau \right], \quad (155)$$

where t and t_0 are the current and initial time instant respectively, \mathbf{K}_I is a positive definite gain matrix. By reasonably assuming that $\boldsymbol{\omega}_R(t_0) = \mathbf{0}_3$, it implies that $\mathbf{q}^R(t_0)$ is null as well. By taking the time derivative of (155), through (154), the following dynamics for the residual vector is obtained

$$\dot{\hat{\boldsymbol{\tau}}}_R^R + \mathbf{K}_I \hat{\boldsymbol{\tau}}_R^R = \mathbf{K}_I \boldsymbol{\tau}_R^R. \quad (156)$$

Equation (156) is a first order low-pass dynamic system: it can be easily recognized that $\hat{\boldsymbol{\tau}}_R^R \rightarrow \boldsymbol{\tau}_R^R$ when $t \rightarrow \infty$ and with $\mathbf{K}_I \simeq \infty$. Thus, by properly choosing the matrix \mathbf{K}_I , it is possible to achieve a good estimation of $\boldsymbol{\tau}_R^R$ while, at the same time, a low pass filtering of the high-frequency noise.

Estimation of the wrench acting on the tool tip. Once both $\hat{\mathbf{f}}_R$ and $\hat{\boldsymbol{\tau}}_R^R$ are known, the estimated wrench acting on the tool tip, $\hat{\mathbf{w}}_E$ is computed as

$$\hat{\mathbf{w}}_E = \mathbf{H}_E^{-T} \begin{bmatrix} \hat{\mathbf{f}}_R \\ \mathbf{R}_R \hat{\boldsymbol{\tau}}_R^R \end{bmatrix}. \quad (157)$$

An illustrative example of the wrench observer's precision is presented in Fig. 115. For the sake of clarity, the figure reports only the second and third component of $\hat{\mathbf{w}}_E$ in (157) (continuous lines) against data of an ATI45 force-torque sensor (dotted lines). As for the first component, it shows a behaviour similar to the second one.

4.2 Admittance Filter

In order to achieve bounded forces exchanged with the environment, a compliant behavior could be enforced between the position and orientation of the end-effector and the interaction generalized forces.

Assigned a planned desired trajectory of the end-effector in terms of position $\mathbf{p}_{E,d}$, orientation $\mathbf{R}_{E,d}$, velocities $\mathbf{v}_{E,d} = [\dot{\mathbf{p}}_{E,d}^T \mathbf{R}_{E,d}^T \boldsymbol{\omega}_{E,d}^T]^T$, and accelerations $\dot{\mathbf{v}}_{E,d}$, the corresponding set of reference motion variables to be fed to the motion controller, $(\mathbf{p}_{E,r}, \mathbf{R}_{E,r}, \mathbf{v}_{E,r}, \dot{\mathbf{v}}_{E,r})$, can be generated via an admittance filter, characterized by the following dynamics

$$\mathbf{M}_E \Delta \dot{\mathbf{v}}_E + \mathbf{D}_E \Delta \mathbf{v}_E + \mathbf{K}_E \mathbf{e}_E = \hat{\mathbf{w}}_E, \quad (158)$$

where $\Delta \mathbf{v}_E = \mathbf{v}_{E,d} - \mathbf{v}_{E,r}$ is the velocity error, while \mathbf{e}_E is the pose error given by

$$\mathbf{e}_E = \left[(\mathbf{p}_{E,d} - \mathbf{p}_{E,r})^\top \left(\frac{1}{2} (\mathbf{R}_{E,d} \mathbf{R}_{E,r}^\top - \mathbf{R}_{E,r} \mathbf{R}_{E,d}^\top)^\vee \right)^\top \right]. \quad (159)$$

The (158) represents the dynamics of a 6-DoF mechanical impedance ([271]) of inertia \mathbf{M}_E , damping \mathbf{D}_E and stiffness \mathbf{K}_E : those matrices are all positive-definite and suitably chosen in a way to impose an over-damped behavior to the system.

Once the reference trajectory of the end-effector has been computed it should be expressed in terms of CoM reference trajectory in order to be tracked by the inner loop pose controller. The reference position and orientation of the robot are then computed as

$$\begin{cases} \mathbf{p}_{R,r} = \mathbf{p}_{E,r} - \mathbf{R}_{R,r} \mathbf{p}_E^R, \\ \mathbf{R}_{R,r} = \mathbf{R}_{E,r} \mathbf{R}_R^E, \end{cases} \quad (160)$$

while the CoM reference velocities and accelerations are obtained by taking the time derivatives of (160). In detail, the reference velocities are given by

$$\begin{cases} \dot{\mathbf{p}}_{R,r} = \dot{\mathbf{p}}_{E,r} - \mathbf{R}_R [\omega_{R,r}]_\times \mathbf{p}_E^R, \\ \omega_{R,r} = \omega_{E,r}, \end{cases} \quad (161)$$

while the reference accelerations are

$$\begin{cases} \ddot{\mathbf{p}}_{R,r} = \ddot{\mathbf{p}}_{E,r} - \mathbf{R}_R [\dot{\omega}_{R,r}]_\times \mathbf{p}_E^R - \mathbf{R}_R [\omega_{R,r}]_\times^2 \mathbf{p}_E^R, \\ \dot{\omega}_{R,r} = \dot{\omega}_{E,r}. \end{cases}$$

5 Experimental Results

5.1 Full-pose Tracking

In Fig. 116 we report the results of an experiment in which the LBF platform shown in Fig. 117 is tasked to track pseudo-sinusoidal trajectory while keeping the orientation horizontal. Thanks to the use of the controller described in Sec. 3, the task is fulfilled as long as the required lateral force is within the bound. When the required force exceeds the bounds the tracking of the orientation is relaxed, however the position tracking is still executed properly. More results and details are provided in [87].

5.2 Interaction Control

In Fig. 118 we report some results of an experiment of interaction control where the platform shown in Fig. 119 is tasked to slide on an inclined surface. The plots

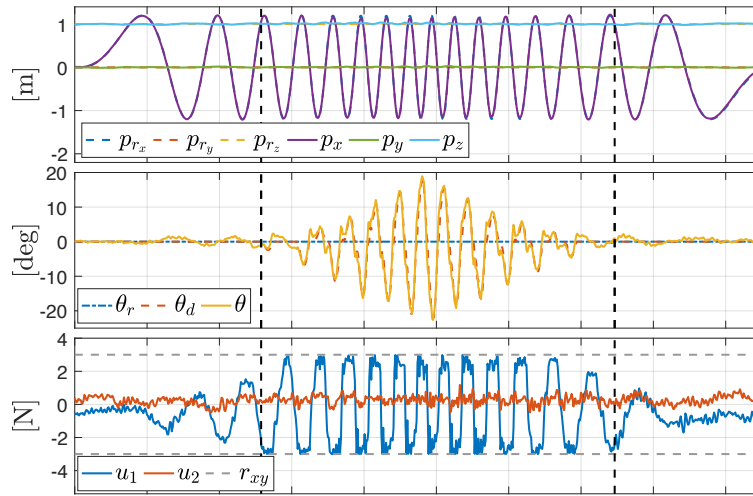


Fig. 116 Exp. 1.1: Desired position: sinusoidal motion along the \mathbf{x}_W axis with constant amplitude and triangular (first increasing then decreasing) frequency. Desired orientation: constantly horizontal. Lateral force bound: constant $r_{xy} = 3\text{ N}$.

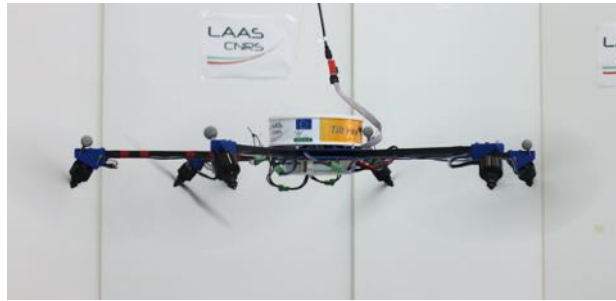


Fig. 117 LAAS-CNRS TiltHex platform used for the tracking experiments.

clearly show how the interaction is stable and interaction forces kept within standard values.

6 Conclusions

In this Chapter we have presented a framework to allow platforms with multi-directional thrust capabilities to perform physical interaction. The method is build around a full-pose tracker, a wrench estimator, an admittance filter and an aerial platform that can accurately execute the total force and moment commands, and measure the real exerted force thanks to the precise control and measurement of the rotational speed of the propellers. It is worth to mention that such platforms and

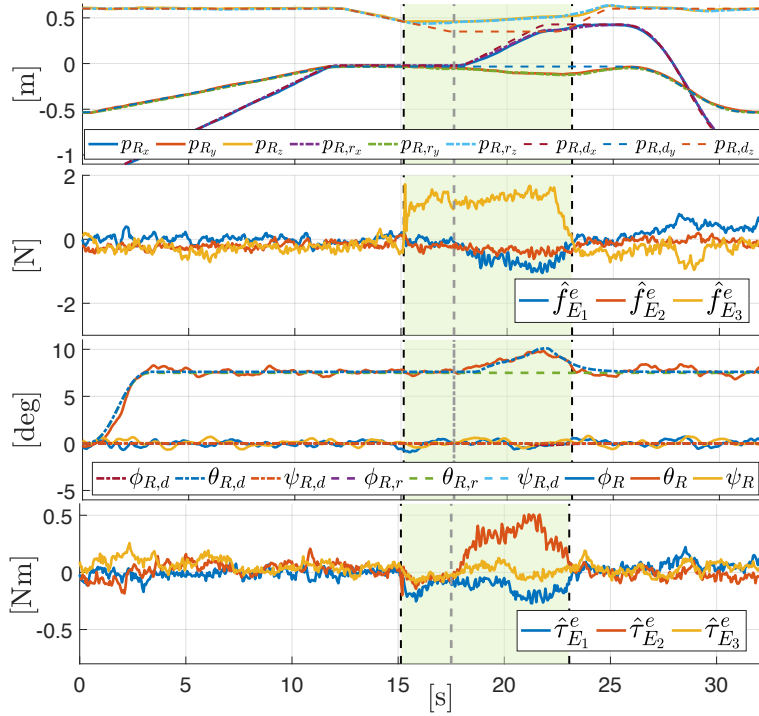


Fig. 118 Sliding with the tool-tip on a tilted surface. The contact phase is highlighted in green, while the actual sliding starts at the gray dashed line: 1) Desired, reference and actual position of the tool-tip. 2) Estimated tool-tip contact forces - low pass filtered (157). 3) Desired, reference and actual tool-tip orientation. 4) Estimated tool-tip contact torques - low pass filtered output of (157).

framework can be used not only in physical interaction with rigid tool but also for aerial manipulation, as recently demonstrated for example in [280].

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Fig. 119 LAAS-CNRS TiltHex platform used with a rigid tool for physical interaction experiments.

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